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> # Prof. Dr. Serkan Dağ
# ME 451 Introduction to Composite Structures
> # File 6.1
# Example on computation of extensional ([A]), coupling ([B]),
and bending ([Dm])
# stiffness matrices for a laminate

> restart :
with(LinearAlgebra) :
> # Enter the number of plies
> n := 3 :
> # Define extensional, coupling, and bending stiffness matrices
> A := Matrix(3) :
  B := Matrix(3) :
  Dm := Matrix(3) :
> # Define ply surface coordinate vector in meters
> h :=  $\begin{bmatrix} -\frac{7.5}{1000} \\ -\frac{2.5}{1000} \\ \frac{2.5}{1000} \\ \frac{7.5}{1000} \end{bmatrix}$  :
> # Define ply angle vector in radians
> theta :=  $\begin{bmatrix} 0 \\ \frac{\text{Pi}}{6} \\ -\frac{\text{Pi}}{4} \end{bmatrix}$  :
> # Define Qbar array
Qbar := Array(1..3, 1..3, 1..n) :
ArrayNumElems(Qbar);

> # Enter properties of the unidirectional lamina
# From Table 2.1 for graphite/epoxy (unit = MPa)
> E1 := 181000 :
  E2 := 10300 :
  nu12 := 0.28 :
  G12 := 7170 :
> # Calculate elements of the compliance matrix for the
unidirectional lamina
> S11 :=  $\frac{1}{E1}$  :

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$$S12 := -\frac{nu12}{E1} :$$

$$S22 := \frac{1}{E2} :$$

$$S66 := \frac{1}{G12} :$$

> # Calculate elements of the reduced stiffness matrix for the unidirectional lamina

> $Q11 := \frac{S22}{S11 \cdot S22 - S12^2} :$

$$Q22 := \frac{S11}{S11 \cdot S22 - S12^2} :$$

$$Q12 := -\frac{S12}{S11 \cdot S22 - S12^2} :$$

$$Q66 := \frac{1}{S66} :$$

> # Calculate elements of transformed reduced stiffness matrix for each angle lamina

Unit = MPa

> for i from 1 by 1 to n
while true do

$$Qbar[1, 1, i] := Q11 \cdot (\cos(\theta[i, 1]))^4 + Q22 \cdot (\sin(\theta[i, 1]))^4 + 2 \cdot (Q12 + 2 \cdot Q66) \cdot (\cos(\theta[i, 1]))^2 \cdot (\sin(\theta[i, 1]))^2 :$$

$$Qbar[1, 2, i] := (Q11 + Q22 - 4 \cdot Q66) \cdot (\sin(\theta[i, 1]))^2 \cdot (\cos(\theta[i, 1]))^2 + Q12 \cdot ((\cos(\theta[i, 1]))^4 + (\sin(\theta[i, 1]))^4) :$$

$$Qbar[1, 3, i] := (Q11 - Q12 - 2 \cdot Q66) \cdot (\sin(\theta[i, 1])) \cdot (\cos(\theta[i, 1]))^3 - (Q22 - Q12 - 2 \cdot Q66) \cdot (\sin(\theta[i, 1]))^3 \cdot \cos(\theta[i, 1]) :$$

$$Qbar[2, 2, i] := Q11 \cdot (\sin(\theta[i, 1]))^4 + Q22 \cdot (\cos(\theta[i, 1]))^4 + 2 \cdot (Q12 + 2 \cdot Q66) \cdot (\cos(\theta[i, 1]))^2 \cdot (\sin(\theta[i, 1]))^2 :$$

$$Qbar[2, 3, i] := (Q11 - Q12 - 2 \cdot Q66) \cdot (\cos(\theta[i, 1])) \cdot (\sin(\theta[i, 1]))^3 - (Q22 - Q12 - 2 \cdot Q66) \cdot (\cos(\theta[i, 1]))^3 \cdot \sin(\theta[i, 1]) :$$

$$Qbar[3, 3, i] := (Q11 + Q22 - 2 \cdot Q12 - 2 \cdot Q66) \cdot (\cos(\theta[i, 1]))^2 \cdot (\sin(\theta[i, 1]))^2 + Q66 \cdot ((\cos(\theta[i, 1]))^4 + (\sin(\theta[i, 1]))^4) :$$

$$Qbar[2, 1, i] := Qbar[1, 2, i] :$$

$$Qbar[3, 1, i] := Qbar[1, 3, i] :$$

$$Qbar[3, 2, i] := Qbar[2, 3, i] :$$

end do:

> # Calculate elements of extensional stiffness matrix [A], coupling stiffness matrix [B], and bending stiffness matrix [Dm]
Units: [A]--> MPa.m; [B]--> MPa.m^2; [Dm]--> MPa.m^3

> for i from 1 by 1 to 3

while true do

for j from 1 by 1 to 3

while true do

$$A[i, j] = 0 :$$

$$B[i, j] := 0 :$$

$$Dm[i, j] := 0 :$$

for k **from** 1 **by** 1 **to** n

while *true* **do**

$A[i,j] := A[i,j] + Qbar[i,j,k] \cdot (h[k+1,1] - h[k,1]) :$

$B[i,j] := B[i,j] + \frac{1}{2} \cdot Qbar[i,j,k] \cdot (h[k+1,1]^2 - h[k,1]^2) :$

$Dm[i,j] := Dm[i,j] + \frac{1}{3} \cdot Qbar[i,j,k] \cdot (h[k+1,1]^3 - h[k,1]^3) :$

end do:

end do:

end do:

> *evalf*(A);

$$\begin{bmatrix} 1739.240863 & 388.3864106 & 56.6337309 \\ 388.3864106 & 453.2535123 & -114.0636096 \\ 56.6337309 & -114.0636096 & 452.4825438 \end{bmatrix}$$

(2)

> *evalf*(B);

$$\begin{bmatrix} -3.128833804 & 0.9855215539 & -1.071656126 \\ 0.9855215539 & 1.157790698 & -1.071656126 \\ -1.071656126 & -1.071656126 & 0.9855215540 \end{bmatrix}$$

(3)

> *evalf*(Dm);

$$\begin{bmatrix} 0.03343203414 & 0.006460977237 & -0.005240293688 \\ 0.006460977237 & 0.009319771318 & -0.005595913147 \\ -0.005240293688 & -0.005595913147 & 0.007662779735 \end{bmatrix}$$

(4)

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